

11/09/2020

UNIVERSITA' DEGLI STUDI DI PADOVA

Corso di Dottorato in Scienze, Tecnologie e Misure Spaziali

Study and Development of a Fluidic System for Iodine-fed Magnetically Enhanced Plasma Thruster (MEPT)

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Framework and Statement of the Problem

MEPT



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The Magnetically Enhanced Plasma Thruster (MEPT) is an innovative low-cost electric propulsion system able to increase small spacecrafts mobility, opening new unconventional mission scenarios.



T4i is engaged in the design and development of a complete propulsion module based on the MEPT. The module is intended for CubeSat platforms ranging in size from **6 U** to **24 U**, providing:

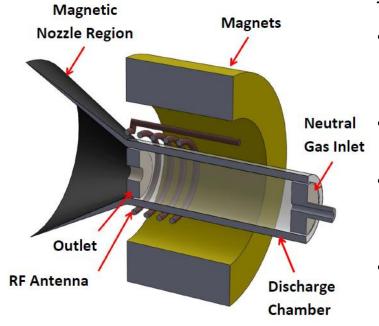
- 0.25-0.65 mN of Thrust
- Isp up to 650 s
- input power lower than 60 W



MEPT



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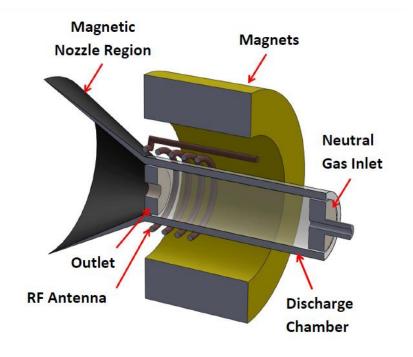
The main components of MEPT are:

- A fluidic line which transfers the neutral gas propellant from a storage tank to the discharge chamber.
- A **discharge chamber** inside which the neutral gas is ionized
- A RF antenna, in the MHz frequency range, which generates the electromagnetic (EM) fields for gas ionization
- Magnets producing a magnetostatic field to enhance the plasma confinement and provide the magnetic nozzle effect.

MEPT



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Advantages:

- Absence of electrodes immersed in the plasma
- Good power scalability
- Adaptability to different propellants
- No need for a neutralizer

Disadvantage:

High thermal load

MEPT: Iodine Propellant

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MEPT can work with different propellants (such as Ar, Kr, Xe, Air, CO2). Because of this last feature it seems extremely promising to investigate the employment of **Iodine as a propellant**, which is particularly appealing for space applications.

Why Iodine Propellant?

- It costs only 1/5 compared to Xenon
- It can be stored as solid
- High density
- No pressurized tank

Disadvantages:

- Chemically reactive
- Never flown before





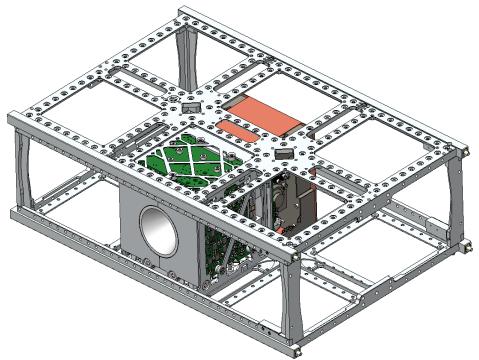
Research Project

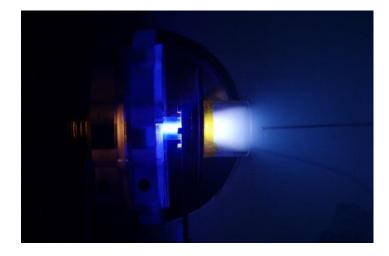
Aim of the Project



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The research program will be focused mainly on the design of an innovative **low cost fluidic system for Iodine fed Magnetically Enhanced Plasma Thruster**, in order to use it on a Cubesat platform.





The fluidic subsystem must provide a **fixed mass flow rate of 0.1 mg/s ±10%** to the thruster.

Research Activity



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- 1. Study and optimization of the mass flow control system by means of thermal management strategy, in order to grant the proper sublimation rate and to avoid the recondensation.
- 2. Development of a proper **software tool** to design and optimize the system from a thermal and fluidic point of view.
- 3. Testing of **the mass flow control system with lodine propellant**.

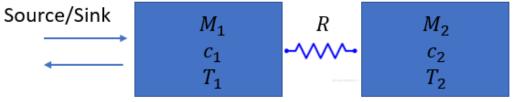


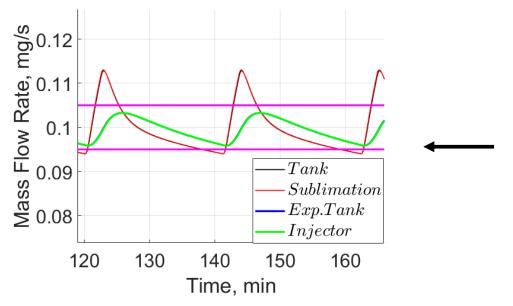
Activities up to now

Numerical Approach



A Thermal Lumped Parameter Model was developed in order to simulate the thermal behaviour of the system. M = mass c = specific heat T = Temperature R = radiative/conductive resistance





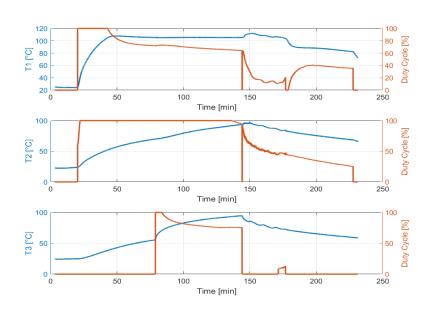
A Fluidic Model, coupled with the thermal one, was developed and was used to study the mass flow control system.

Tests

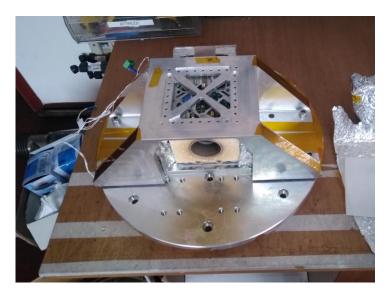


The following tests have been performed:

- Tuning of the thermal control loop
- Calibration of the sensors
- Functional tests in vacuum with the thruster module (heating, ignition, thrusting and cooling)
- TVAC





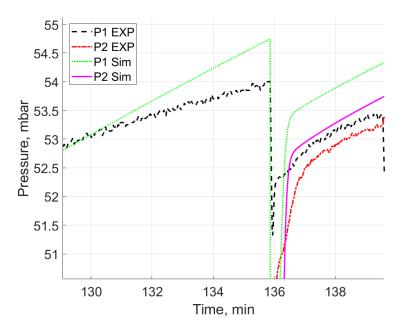


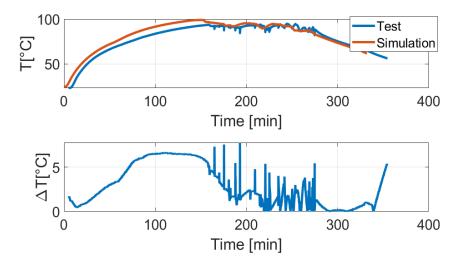
Validation of the Numerical Model



The following steps were followed:

- 1. Calibration of the thermal resistances
- 2. Validation of the temperature profile vs time





- 3. Calibration of the fluidic parameters
- 4. Validation of the pressure profile vs time



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Future Work

- 1. To finish the fluidic tests;
- 2. To conclude the optimization of the fluidic module.
- 3. To write the thesis









Summary of Activities

- ✓ Bibliography Research
 - ✓ International Papers
- ✓ Numerical Models
- ✓ Thermal Model
- Fluidic Model
- ✓ Coupling
- Validation
- Design and Development
 - Mass Flow Control System
 - Optimization
- Calibration and Test
- Thermal Control
- Mass Flow Control

Legend

- ✓ Finished
- In progress
- To start



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Thanks for your attention...

... any questions?