12/09/2019

UNIVERSITA' DEGLI STUDI DI PADOVA

Corso di Dottorato in Scienze, Tecnologie e Misure Spaziali

Study and Development of a Fluidic System for lodine-fed Magnetically Enhanced Plasma Thruster (MEPT)

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Framework and Statement of the Problem

MEPT



The Magnetically Enhanced Plasma Thruster (MEPT) is an innovative low-cost electric propulsion system able to increase small spacecrafts mobility, opening new unconventional mission scenarios.



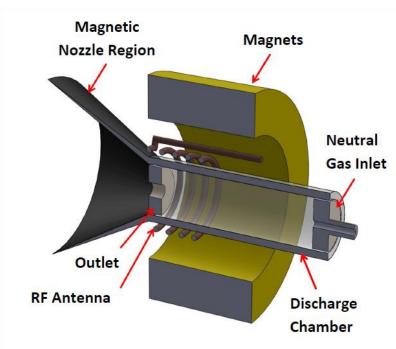
T4i is engaged in the design and development of a complete propulsion module based on the MEPT. The module is intended for CubeSat platforms ranging in size from **6 U** to **24 U**, providing:

- 0.01-1 mN of Thrust
- Isp up to 850 s
- input power lower than 80 W



MEPT



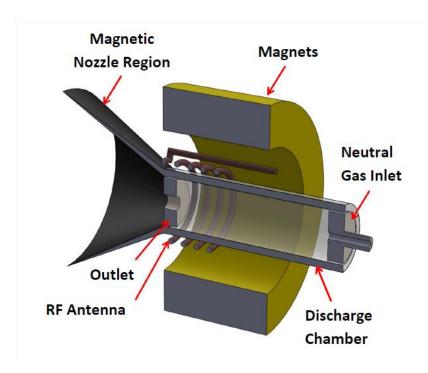


The main components of MEPT are:

- A fluidic line which transfers the neutral gas propellant from a storage tank to the discharge chamber.
- A dielectric tube inside which the neutral gas is ionized
- A RF antenna, in the MHz frequency range, which generates the electromagnetic (EM) fields for gas ionization
- Magnets producing a magnetostatic field to enhance the plasma confinement and provide the magnetic nozzle effect.

MEPT





Advantages:

- Absence of electrodes immersed in the plasma
- Good power scalability
- Adaptability to different propellants
- No need for a neutralizer

Disadvantage:

 High thermal load due to plasma

MEPT: Iodine Propellant



MEPT can work with different propellants (such as Ar, Kr, Xe, Air, CO2). Because of this last feature it seems extremely promising to investigate the employment of **Iodine as propellant**, which is particularly appealing for space applications.

Why Iodine Propellant?

- It costs only 1/5 compared to Xenon
- It can be stored as solid
- High density
- No pressurized tank

Disadvantages:

- Chemically reactive
- Never flown before



On Similar System

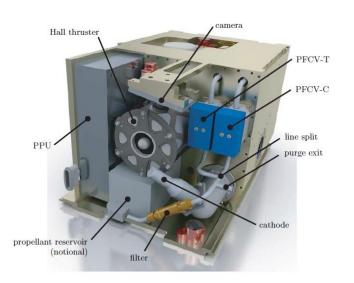


Recently, some companies have implemented a lodine Fed-System for Hall Effect Thruster or Ion Thruster.

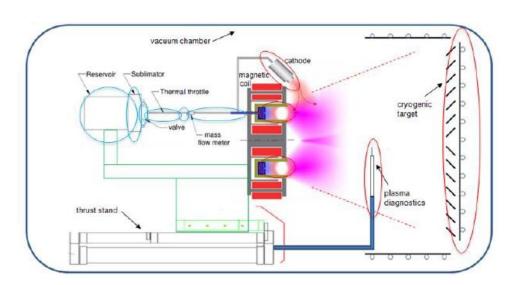
Busek – Ion Thruster



iSAT/NASA – Hall Thruster



SITAEL - Hall Thruster





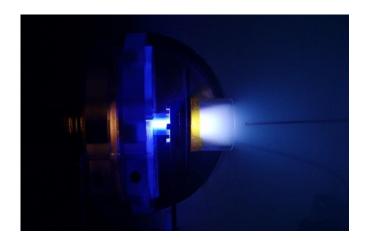
Research Project

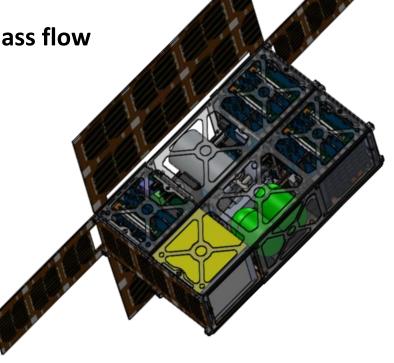
Aim of the Project



The research program will be focused mainly on the design of an innovative low cost fluidic system for lodine fed Magnetically Enhanced Plasma Thruster, in order to use it on a Cubesat platform.

The fluidic subsystem must provide a fixed mass flow rate of 0.1 mg/s ±10% to the thruster.





Research Activity



- 1. Development and testing of an innovative low cost **Mass**Flow Meter for Iodine fed MEPT.
- 2. Development of the mass flow control system by means of thermal management strategy, in order to grant the proper sublimation rate and to avoid the recondensation.
- 3. Testing of the mass flow control system with lodine propellant.

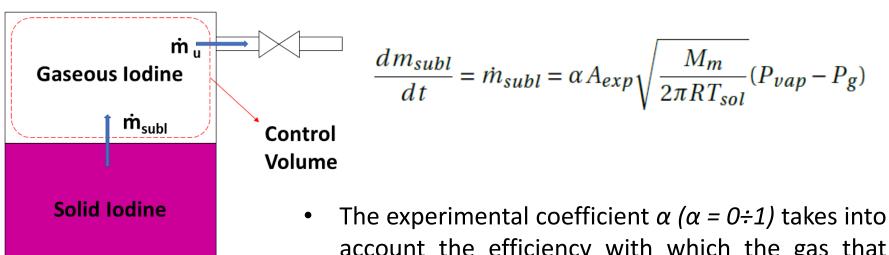


Activities up to now

Investigation of physics



Sublimation



- account the efficiency with which the gas that comes in contact with the solid surface condense
- A_{exp} represents the sublimation surface of the solid at the temperature T_{sol}

Numerical Approach

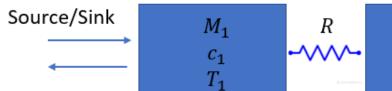


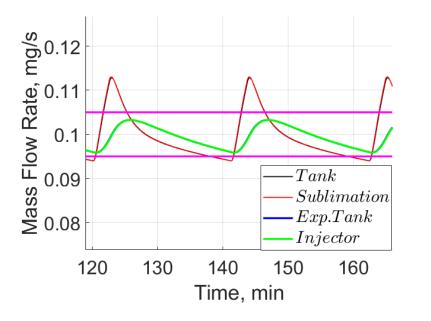
 M_2

 c_2

 T_2

A Thermal Lumped Parameter Model was developed in order to simulate the thermal behaviour of the system. M = mass c = specific heat T = Temperature R = radiative/conductive resistance





A Fluidic Model, coupled with the thermal one, was developed and is used to design the mass flow control system.



Future Work

Future Work



- 1. To set up the experimental apparatus;
- 2. To test and calibrate the Mass flow meter.
- 3. To test and characterize the thermal and mass flow control system.







Summary of Activities



- ✓ Bibliography Research
- ✓ International Paper
- ✓ Numerical Models
- ✓ Thermal Model
- ✓ Fluidic Model
- Design and Development
 - Mass Flow Meter
 - Mass Flow Control System
- Calibration and Test
- ☐ Mass Flow Meter
- ☐ Mass Flow Control System

Legend

- ✓ Finished
- In progress
- ☐ To start



Thanks for your attention...

... any questions?