

Numerical and Experimental Investigation into the Performance of Plasma Sources for Space Propulsion Systems

Mirko Magarotto

Centro di Ateneo di Studi e Attività Spaziali Giuseppe Colombo
CISAS
University of Padova

Helicon
Plasma
Sources

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Framework
& Statement
of the
Problem

Numerical
Approach

Experimental
Approach

Future
Expected
Results

- 1 Framework & Statement of the Problem
- 2 Numerical Approach
- 3 Experimental Approach
- 4 Future Expected Results

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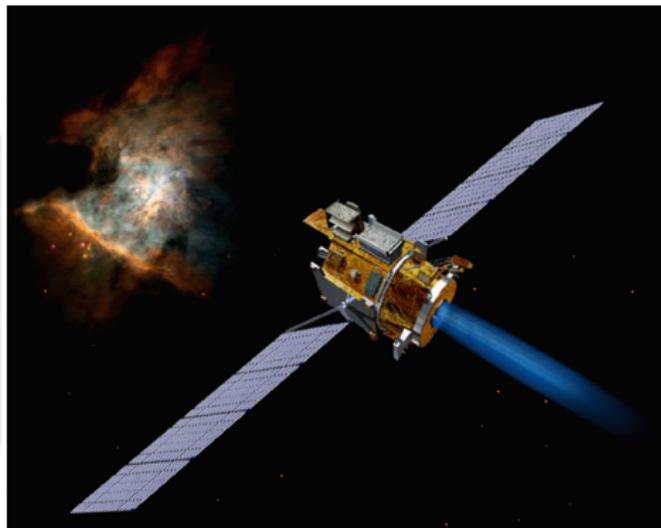
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Definition

- Electric power employed to generate thrust
- Usually plasma is operation fluid



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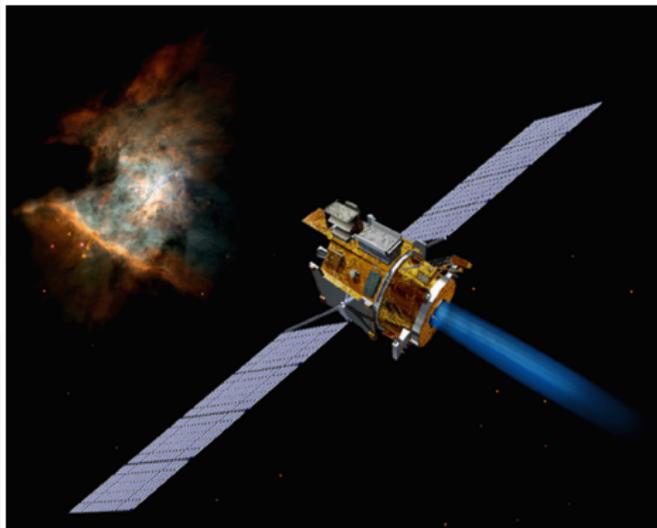
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Main Features

- High specific impulse:
higher > 1000 s
- Low thrust:
lower < 1 N



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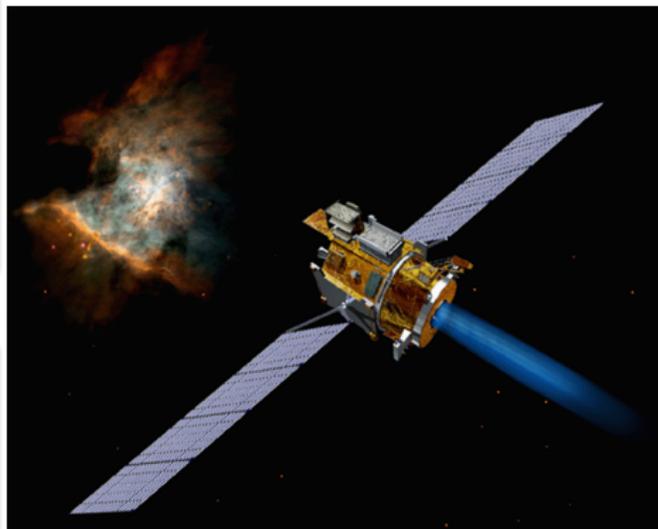
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Some applications

- Attitude control
- Cubesats
- Interplanetary missions



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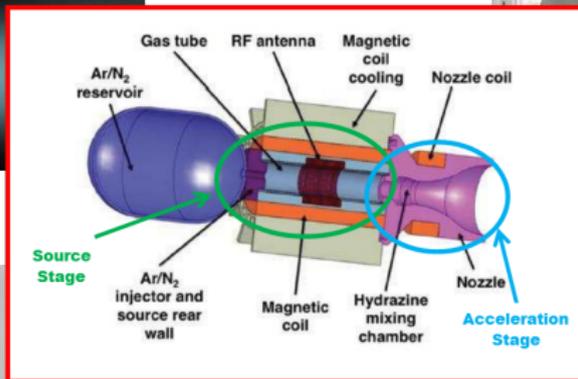
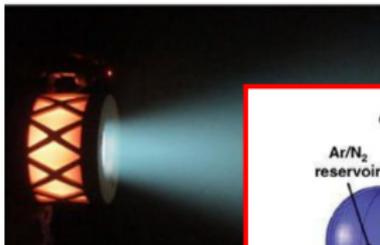
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Electric Space Propulsion - State of the art

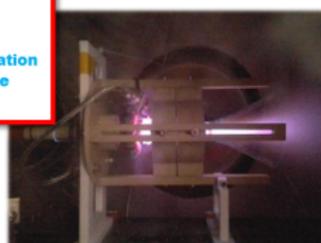
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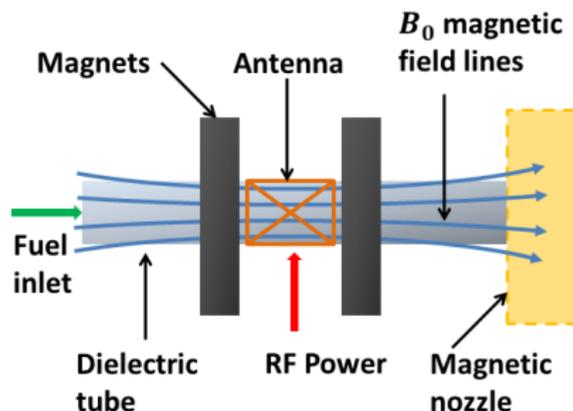
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Experimental
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Expected
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Main components

- Dielectric Tube
- RF Antenna
- Magnets

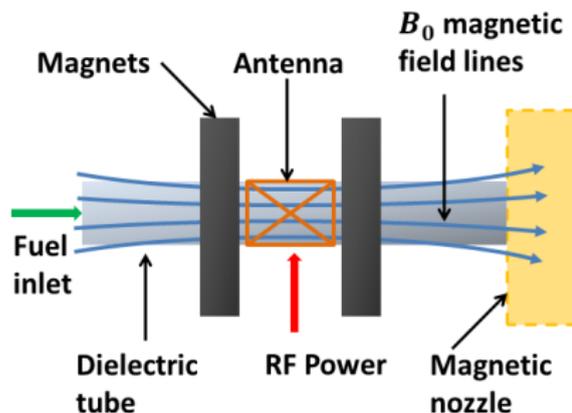


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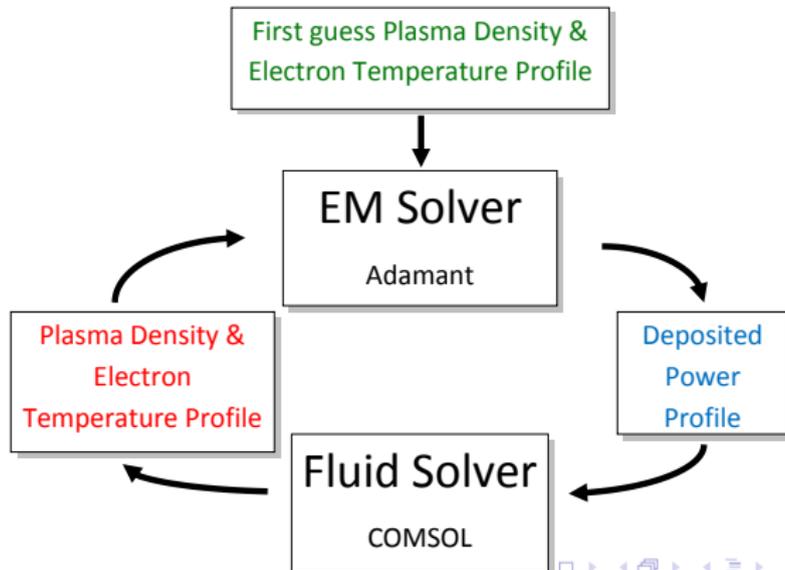
Main features

- high density plasma
 $n \geq 10^{18} \text{ m}^{-3}$
- simple antenna geometry
- low magnetic field
 $B_0 \leq 1000 \text{ G}$



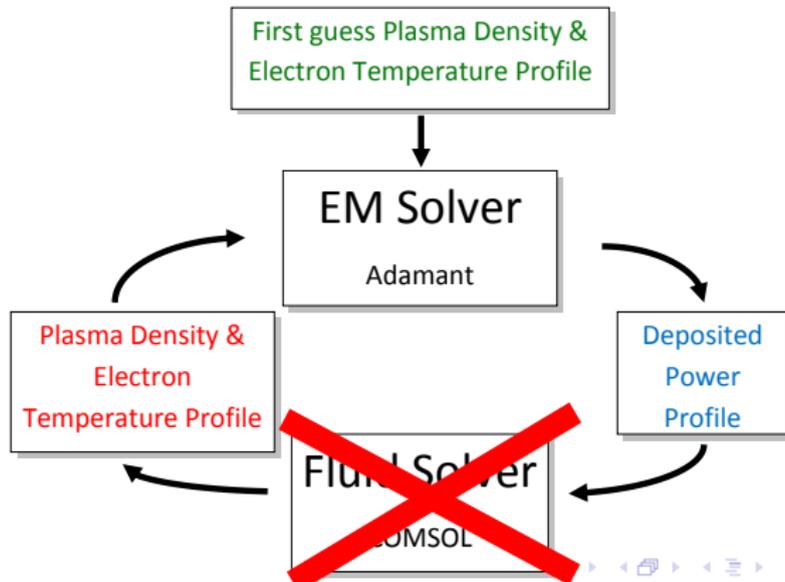
Coupling FLUID SOLVER and EM SOLVER

- Reproduction of the transport for high-density plasma
- Computational cost at bay



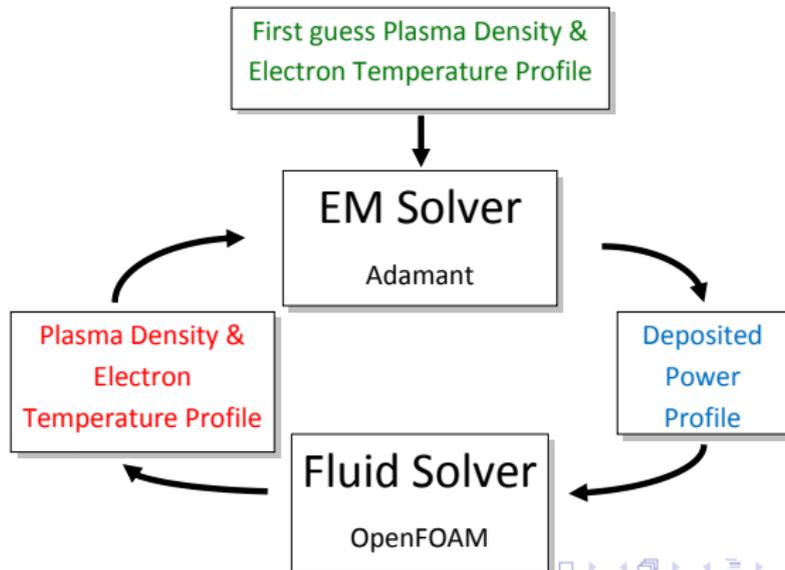
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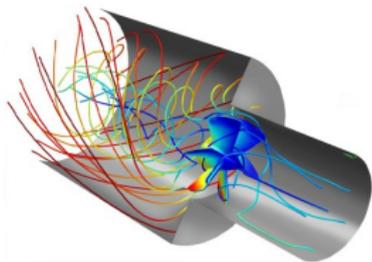
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COMSOL

- Commercial software
- Limited possibility of modifying the model's equations
- Problems with energy equation and boundary conditions



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Experimental
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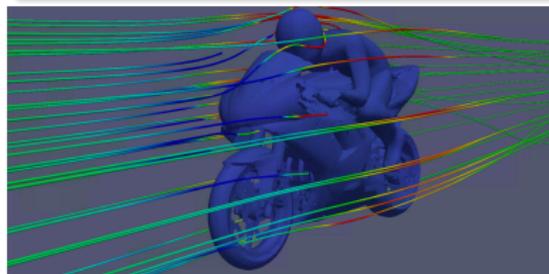
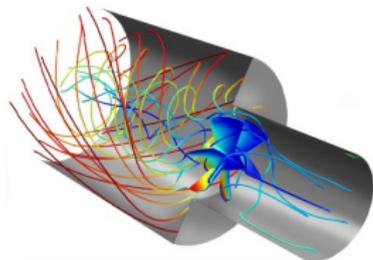
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OpenFOAM

- Open source C++ library
- Total access to the source code, and to model's equations
- Energy equation and boundary conditions written by user



Implementation Strategy

- 1 Implementation in OpenFOAM of the same fluid model of COMSOL, in order to have a consolidated benchmark

Implementation Strategy

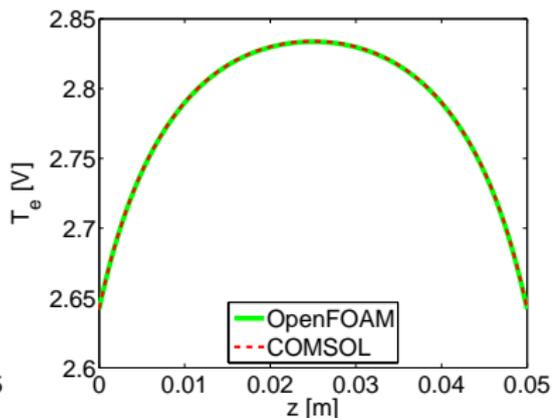
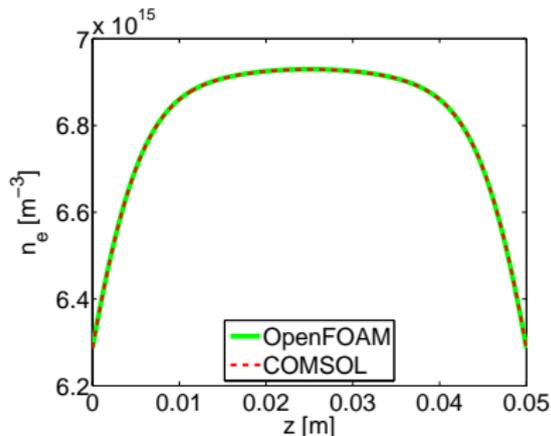
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- 3 Update of OpenFOAM with more refined fluid model (e.g. modified energy equation)
- 4 Experimental validation



1D Simulation

- Evaluated plasma parameters gradients only along axis of cylindrical plasma source
- Assumed power deposition profile
- Comparison between the two solvers very good

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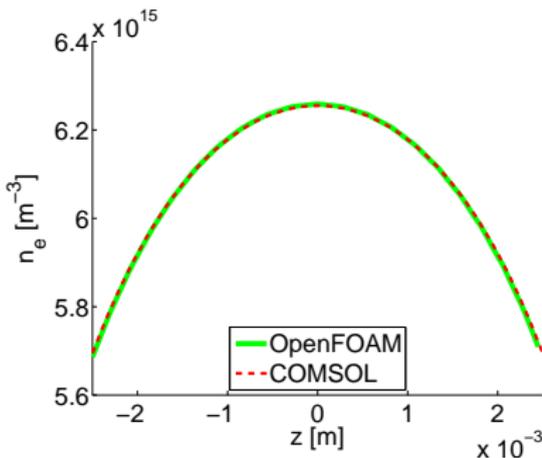
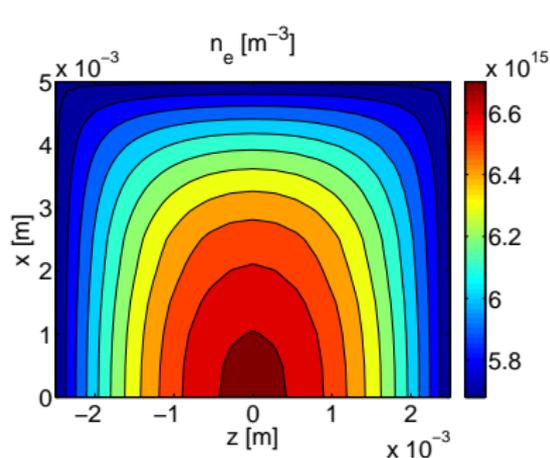
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Numerical
Approach

Experimental
Approach

Future
Expected
Results



3D Simulation

- 3D data sampled on a semi-plane which contains the axis of the plasma source, and on a line into this plane
- Assumed power deposition profile
- Comparison between the two solvers very good

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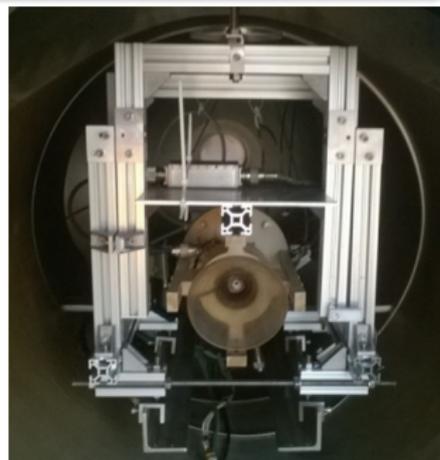
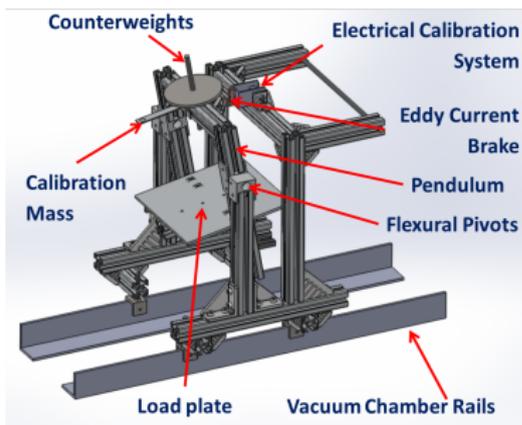
Numerical
Approach

Experimental
Approach

Future
Expected
Results

Thruster Diagnostic System

Faraday probe, Retarding Potential Analyzer and Thrust Balance to measure Specific Impulse and Thrust



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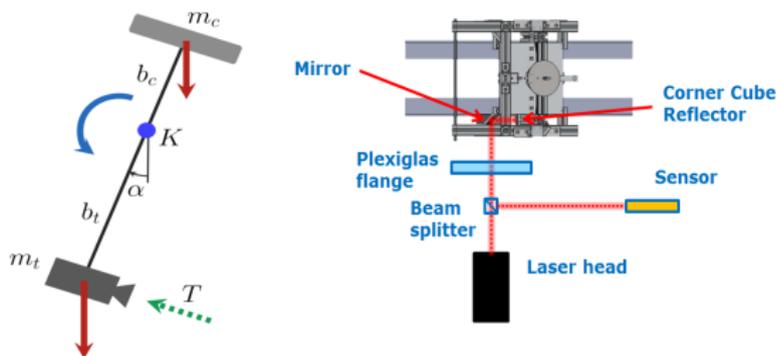
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Framework
& Statement
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Problem

Numerical
Approach

Experimental
Approach

Future
Expected
Results



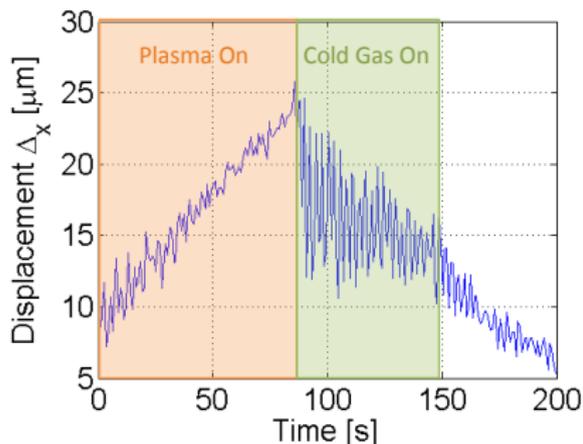
Counterbalanced Pendulum Concept

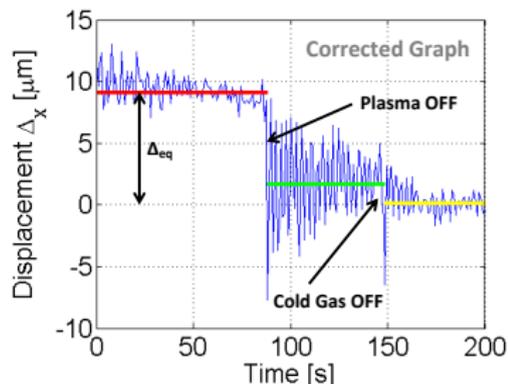
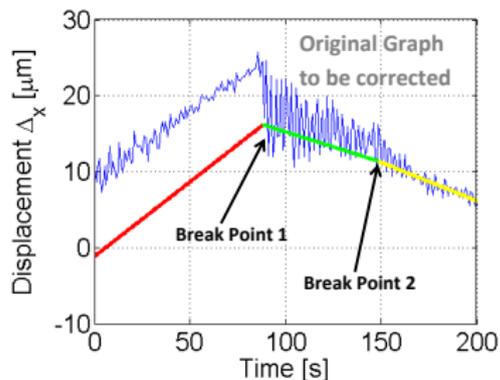
- In accordance with the rotational equilibrium equation a thrust T produces an angular displacement:
$$\alpha = Tb_t / (K + g(m_t b_t - m_c b_c))$$
- Measured, with a laser interferometer, the displacement of a corner cube fastened to the pendulum arm

Position Drift

Due to thermal gradients which make the pendulum mass center move

- Plasma heat losses major drift source
- Electrical cables and gas adduction tube are other important drift sources





Drift Correction Procedure & Thrust Evaluation

- Identification of intervals where heating conditions uniform
- Drift contribution approximated with best fit lines
- Thrust evaluated from Δ_{eq} , corrected mean values difference

Test	1	2	3	4	5
Thrust [mN]	0.278	0.426	0.380	0.252	0.405
Uncertainty [mN]	± 0.020	± 0.023	± 0.047	± 0.024	± 0.054

Thrust Measurement

Non optimized 50 W Helicon Plasma Thruster, uncertainty in the order of 10%

Thrust Stand Results

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Test	1	2	3	4
T electrical [mN]	0.203	0.254	0.147	0.180
T stand [mN]	0.178	0.208	0.172	0.192
Relative Difference [%]	-12.4	-18.2	16.6	6.8

Stand VS Electrical Measurements

Agreement within the 20%, in line with the Electrical measurements uncertainty in the order of 30-40%

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- 2 Design, optimization, and testing of an high-power Helicon plasma source

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- 1 Full development of the numerical tool devoted to the source analysis
- 2 Design, optimization, and testing of an high-power Helicon plasma source
- 3 Technology exploitation



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Numerical
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Experimental
Approach

Future
Expected
Results